

**AMENDMENTS TO THE CLAIMS**

**Please amend claims 1-2 and 4-7 and cancel claims 3 and 8-14 as follows.**

1. (Currently Amended) A gas turbine engine comprising ~~rotary members~~ turbine blades which rotate together with a rotary shaft and a ~~easing~~ shroud which surrounds said ~~rotary members~~ turbine blades,

wherein no hard abrasive layer is provided on said turbine blade for said shroud,

wherein said easing shroud has, on a surface thereof, a coating an abradable layer which has a hardness lower than that of a material of said ~~rotary members~~ turbine blades, and said abradable layer is formed on a surface of said shroud, and

wherein said abradable layer is formed by thermally spraying only a raw material powder sieved to have a particle size not more than 125  $\mu$ m by a high velocity oxygen-fuel method so as to have said coating has a porosity of 5% to 30% by volume and an oxygen content of not more than 2% by weight.

2. (Currently Amended) The gas turbine engine according to claim 1, wherein said ~~coating~~ abradable layer is composed of an alloy of M-Cr-Al-Y provided that M represents Ni, Co, or Ni-Co.

3. (Canceled)

4. (Currently Amended) The gas turbine engine according to claim 1, further comprising an intermediate layer which is disposed between said ~~easing~~ shroud and said ~~coating~~ abradable layer in order to absorb a difference in coefficient of thermal expansion.

5. (Currently Amended) The gas turbine engine according to claim 1, wherein said porosity of said ~~coating~~ abradable layer is 10% by volume.

6. (Currently Amended) The gas turbine engine according to claim 1, wherein said ~~coating~~ abradable layer is made of stabilized zirconia.

7. (Currently Amended) The gas turbine engine according to claim 2, wherein said ~~coating~~ abratable layer is composed of an alloy of Co-32%Ni-21%Cr-7.5%Al-0.5%Y provided that numerals indicate percent by weight.

8-14. (Canceled)

**Please add new claims 15 and 16 as follows.**

15. (New) The gas turbine engine according to claim 1, wherein said abratable layer has a thickness of 0.5 to 2.0 mm.

16. (New) The gas turbine engine according to claim 1, wherein said raw material powder is sieved to have a particle size not more than 75  $\mu\text{m}$ .